

**AMENDMENTS TO THE CLAIMS:**

This listing of claims will replace all prior versions, and listings, of claims in the application:

1. (Previously Presented) A combustor liner for a gas turbine, the combustor liner having a substantially cylindrical shape; and a plurality of axially spaced annular grooves formed in an outside surface of said combustor liner, each groove having a uniform cross-section and extending continuously about a circumference of said liner.
2. (Original) The combustor liner of claim 1 wherein said grooves are substantially semi-circular in cross-section.
3. (Original) The combustor liner of claim 1 wherein said grooves are arranged transversely to a direction of cooling air flow.
4. (Original) The combustor liner of claim 1 wherein said grooves are semi-circular in cross-section, and have a diameter  $D$ , and wherein a depth of said grooves is equal to about 0.05 to 0.50 $D$ .
5. (Original) The combustor liner of claim 4 wherein a center-to-center distance between adjacent grooves is equal to about 1.5-4 $D$ .

6. (Original) The combustor liner of claim 1 wherein a center-to-center distance between adjacent grooves is equal to about  $1.5-4D$ .

7. (Cancelled).

8. (Original) The combustor liner of claim 1 wherein said grooves are angled relative to a direction of cooling air.

9. (Currently Amended) A combustor for a gas turbine, the combustor including a liner having a substantially cylindrical shape; a flow sleeve surrounding said liner; a first plurality of axially spaced, continuous circumferential grooves formed in an outside surface of said liner, angled relative to a direction of cooling air flowing between said liner and said flow sleeve; and a second plurality of axially spaced, continuous circumferential grooves criss-crossed with said first plurality of axially spaced circumferential grooves wherein said first and second plurality of axially spaced circumferential grooves are ~~smoothly~~ uniformly curved in cross-section.

10. (Previously Presented) A combustor liner for a gas turbine, the combustor liner having a substantially cylindrical shape; and a plurality of axially spaced annular grooves formed in an outside surface of said combustor liner, each groove extending continuously about a circumference of said liner; wherein said grooves are semi-circular in cross-section, based on a diameter  $D$ , and wherein a depth of said grooves is equal to about  $0.05$  to  $0.50D$ .

11. (Original) The combustor liner of claim 10 wherein a center-to-center distance between adjacent grooves is equal to about 1.5-4D.

12. (Original) The combustor liner of claim 10 wherein said grooves are substantially semi-circular in cross-section.

13. (Original) The combustor liner of claim 12 wherein a center-to-center distance between adjacent grooves is equal to about 1.5-4D.

14. (Original) The combustor liner of claim 10 wherein said grooves are arranged transversely to a direction of cooling air flow.

15. (Original) The combustor liner of claim 10 wherein said grooves are angled relative to a direction of cooling air flow.

16. (Currently Amended) A combustor liner for a gas turbine, the combustor including a liner having a substantially cylindrical shape; a first plurality of axially spaced, continuous circumferential grooves formed in an outside surface of said liner, angled relative to a direction of cooling air flow; and a second plurality of axially spaced, continuous circumferential grooves criss-crossed with said first plurality of axially spaced circumferential grooves, and wherein said first and second plurality of axially spaced circumferential grooves are smoothly curved in cross-section.